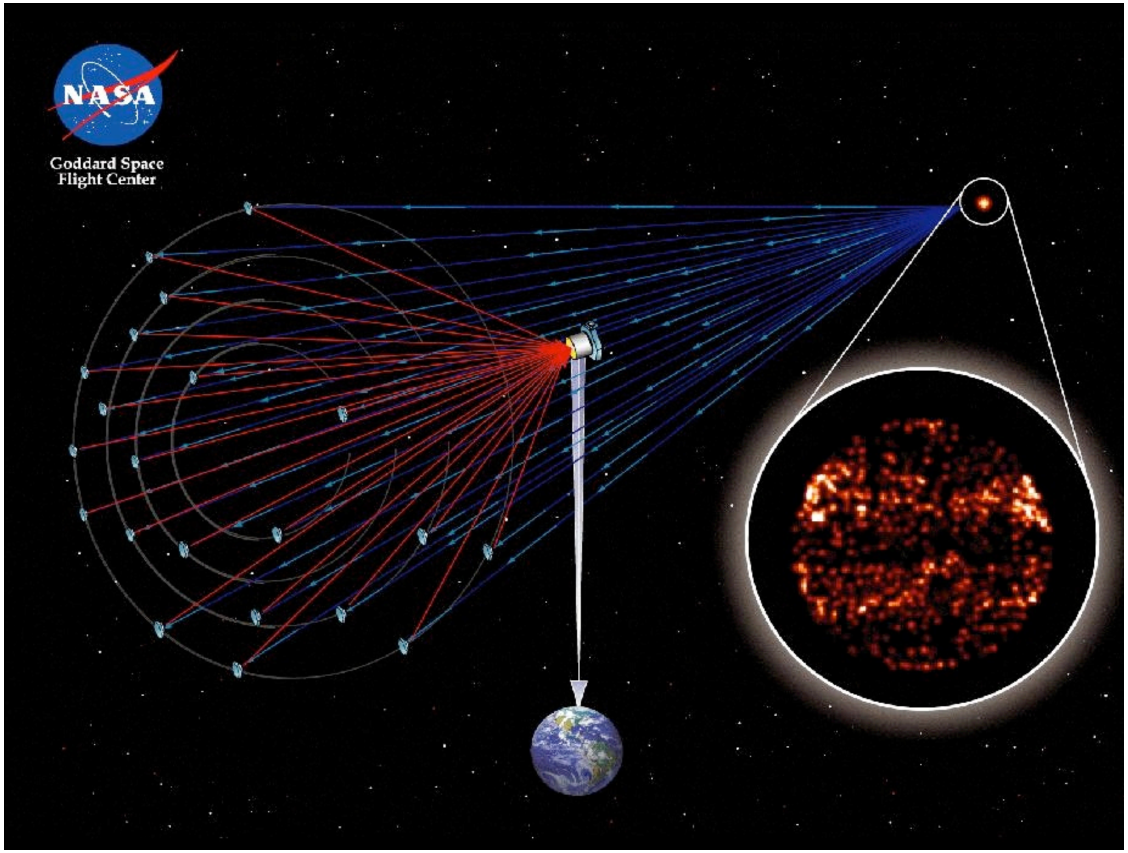
Stellar Imager (SI):
Viewing the UV/Optical Universe in High Definition



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D. Mozurkewich (Seabrook Eng.), and the SI Mission Concept Development Team

URL: http://hires.gsfc.nasa.gov/si/

Presented at the Ares V Workshop at NASA Ames Research Center, April 26-27, 2008, Moffett Field, CA

SI is a space-based, UV/Optical Interferometer (UVOI) with over 200x the resolution of HST

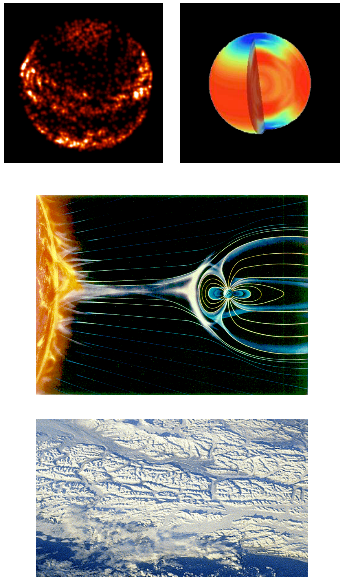
- It will enable 0.1 milli-arcsecond (mas) spectral imaging of stellar surfaces and of the Universe in general

- and

- Open an enormous new "discovery space" for Astrophysics in the UV/Optical with its combination of high (sub-mas) angular resolution, dynamic imaging, and spectral energy resolution

Science goals of the Stellar Imager (1)

- ***Solar and Stellar Magnetic Activity*** **and their impact on *Space Weather, Planetary Climates, and Life***
  - Understand the dynamo process responsible for magnetic activity
  - Enable improved forecasting of solar/stellar magnetic activity on time scales of days to centuries
  - Understand the impact of stellar magnetic activity on planetary climates and on the origin and continued existence of life
  - Techniques:
    - spatially resolving stellar disks to map the evolving atmospheric activity as a tracer of dynamo patterns
    - disk-resolved high temporal resolution asteroseismic probing of internal stellar structure and flows (at least to degrees of order 60)
- ***Magnetic Accretion Processes*** **and their roles in the *Origin & Evolution of Structure* and in the *Transport of Matter* throughout the Universe**.
  - Understand accretion mechanisms in sources ranging from planet-forming systems to black holes
  - Understand the dynamical flow of material and the role of accretion in evolution, structure, and transport of matter in complex interacting systems

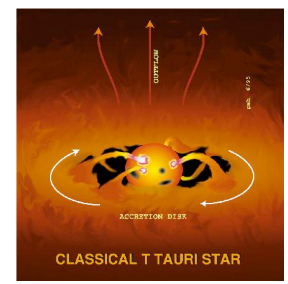


Stellar Activity:
Space Weather
Planetary Climates
Life

Science goals of the Stellar Imager (2)

- **AGN Structure**
  - Understand the close-in structure of AGN including jet forming regions, winds, and transition regions between Broad and Narrow Line Emitting Regions
- **Dynamic Imaging of the Universe at Ultra-High Angular Resolution**
  - Understand the dynamical structure and physical processes in many currently unresolved sources, such as: AGN, SN, PN, interacting binaries, stellar winds and pulsations, forming stars and disks, and evolved stars
- **The study of exo-solar planets by imaging:**
  - transits across stellar disks
  - debris and shells surrounding infant star-disk systems
  - dynamic accretion, magnetic field structure, and star/disk interactions in these systems.

AGN Morphology

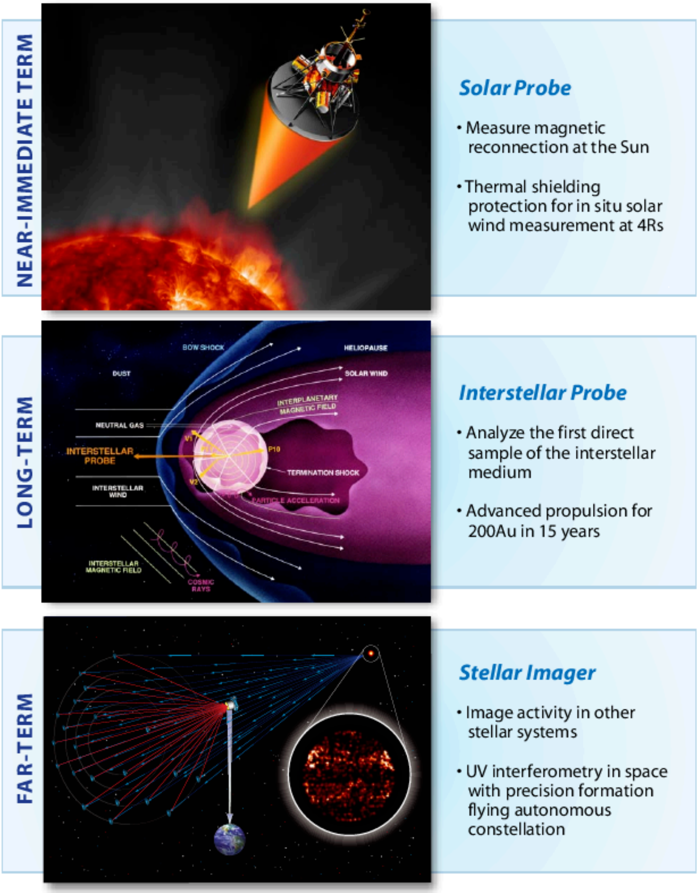


Star/Planet Formation

Stellar Imager is a cross-theme mission addressing Science Goals of both the NASA *Heliophysics* and *Astronomy and Physics* Divisions

- In the Long-Term NASA Strategic Plan, SI is a:
  - “Flagship and Landmark Discovery Mission” in the 2005 Heliophysics Roadmap
  - Potential implementation of the UVOI in the 2006 Science Program for the Astronomy and Physics Division.
  - Candidate Large Class Strategic Mission for the mid-2020's.

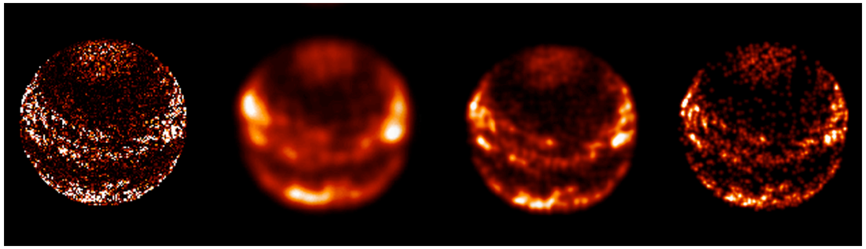
Heliophysics Division Landmark Discovery Missions



Spectral Imaging Capabilities of Stellar Imager

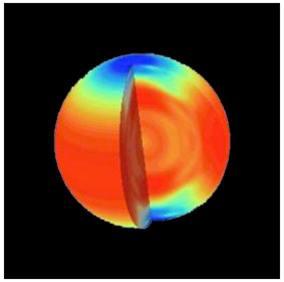
Solar-type star at *4 pc* in *CIV line*

Model ***SIsim* images**



Baseline: 125m 250m 500 m

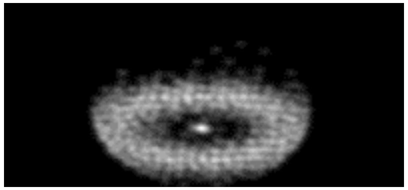
Asteroseismic mapping of internal structure, rotation and flows



Resolution requirements:

- **~20,000km in depth**
- **modes of degree 60 or higher**
- **~1 min. integration times**

Planet formation: magnetosphere-disk interactions

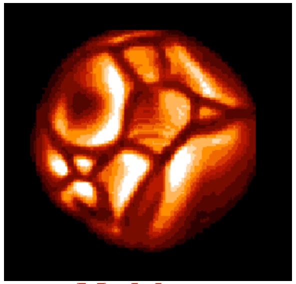


0.1 mas

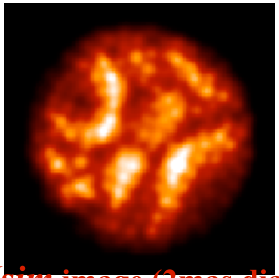
***SI* simulation in Ly a–fluoresced H2 lines**

Baseline: 500 m

Evolved supergiant star at *2 Kpc* in *Mg H&K line*

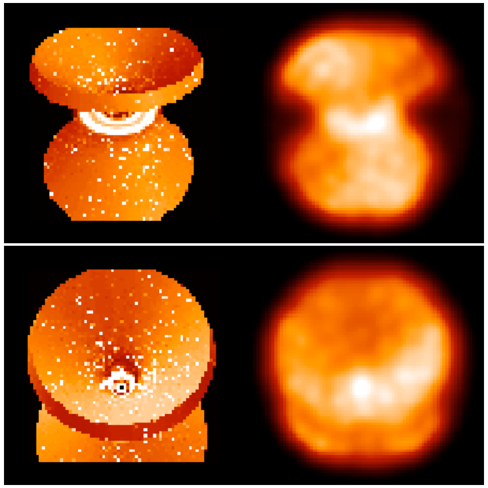


Model



***SIsim* image (2mas dia)**

Imaging of nearby AGN will differentiate between possible BELR geometries & inclinations



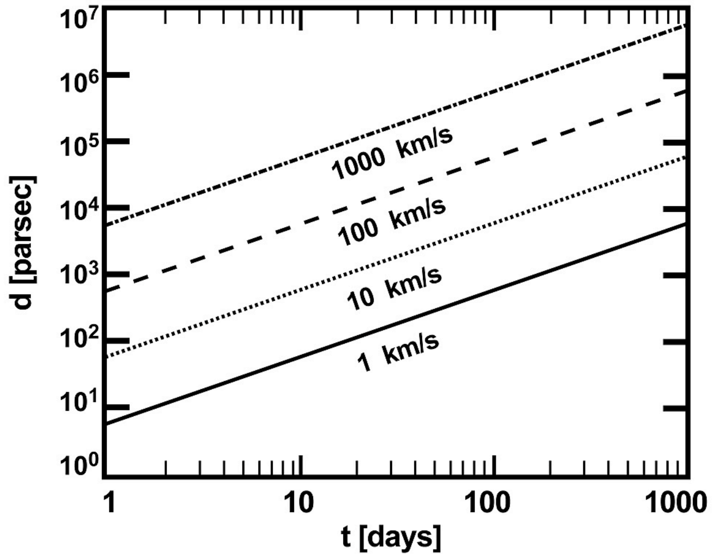
0.1 mas

model ***SI* simulations in *CIV line* (500 m baseline)**

SI will bring the study of the dynamical evolution of many astrophysical objects into reach for the first time

Hours to weeks between successive images will detect dramatic changes in many objects – for example:

- mass transfer in binaries
- pulsation-driven surface brightness variation and convective cell structure in giants and supergiants
- jet formation and propagation in young planetary systems
- reverberating AGN
- and many other variable and evolving sources



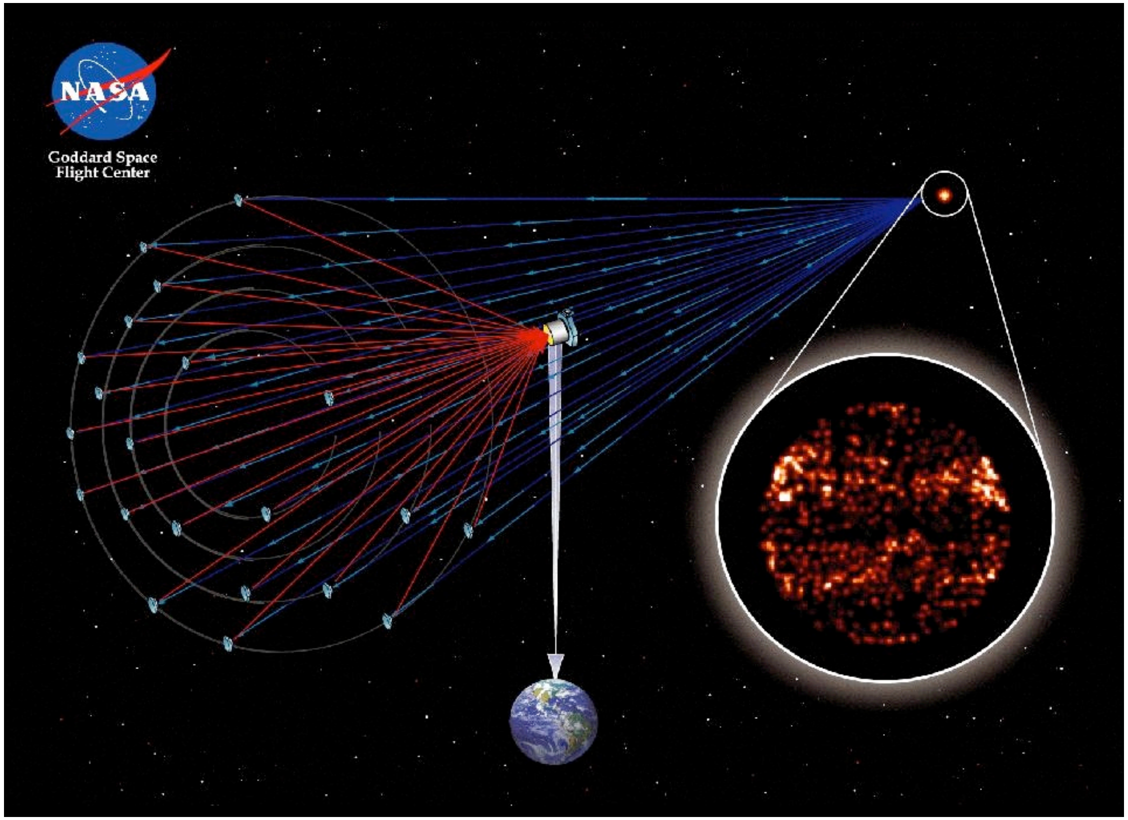
Required Capabilities for SI

- **wavelength coverage: 1200 – 6600 Å**
- **access to UV emission lines** from Lyα 1216 Å to Mg II 2800 Å
  - Important diagnostics of most abundant elements
  - much higher contrast between magnetic structures and background
  - smaller baselines (UV save 2-4x vs. optical, active regions 5x larger)
  - ~10-Å UV pass bands, e.g. C IV (100,000 K); Mg II h&k (10,000 K)
- **broadband, near-UV or optical** (3,000-10,000 K) for high temporal resolution spatially-resolved asteroseismology to resolve internal stellar structure
- angular resolution of 50 μas at 1200 Å (120 μas @2800 Å) to provide ~1000 pixels of resolution over the surface of nearby (4pc) dwarf stars, and more distant giant and supergiant stars.
- **angular resolution of 100 μas in far-UV** for observations of sizes & geometries of AGN engines, accretion processes in forming exo-solar systems, interacting binaries and black hole environs, and for dynamic imaging of evolving structures in supernova, planetary nebulae, AGN, etc.
- **energy resolution/spectroscopy** of R>100 (min) up to R=10000 (goal)
- Selectable "interferometric" and "light bucket/spectroscopic" modes
- a **long-term (~ 10 year) mission**, to enable study of stellar activity cycles:
  - individual telescopes/hub(s) can be refurbished or replaced

SI Concept from Vision Mission (VM) Study

- a 0.5 km diameter space-based UV-optical Fizeau Interferometer
- located near Sun-earth L2 to enable precision formation flying
- 30 primary mirror elements focusing on beam-combining hub
- large advantages to flying more than 1 hub:
  - critical-path redundancy & major observing efficiency improvements

SI Concept from Vision Mission (VM) Study



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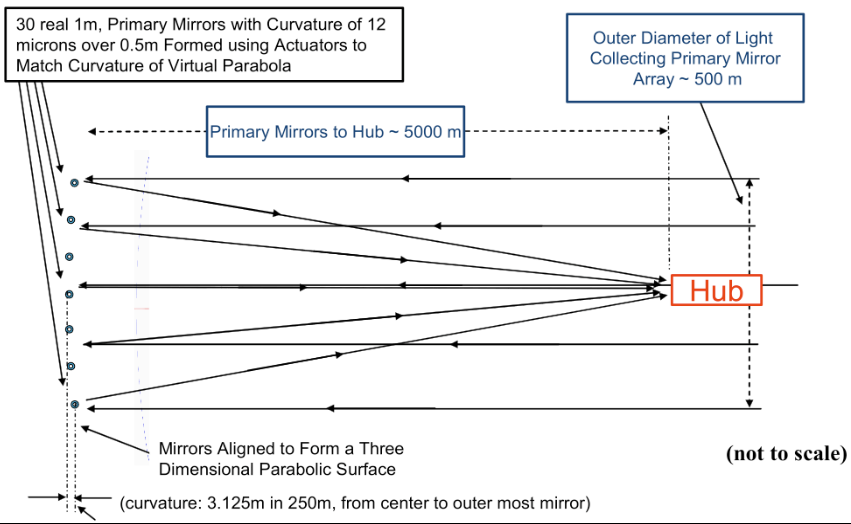
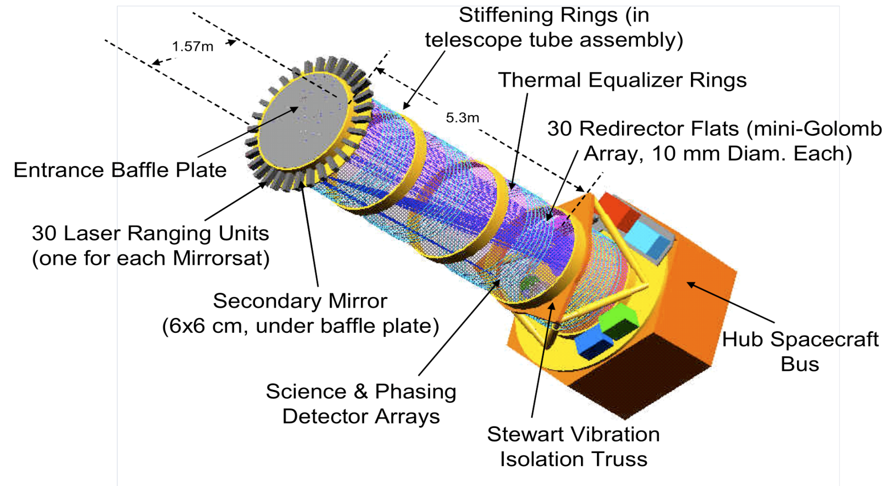
Overview of the VM SI Design Concept

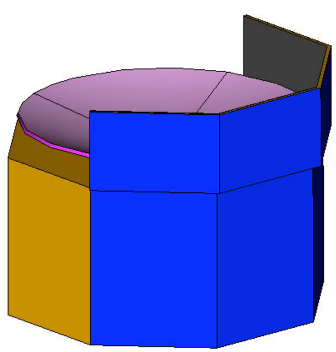
SI Cross-Sectional Schematic



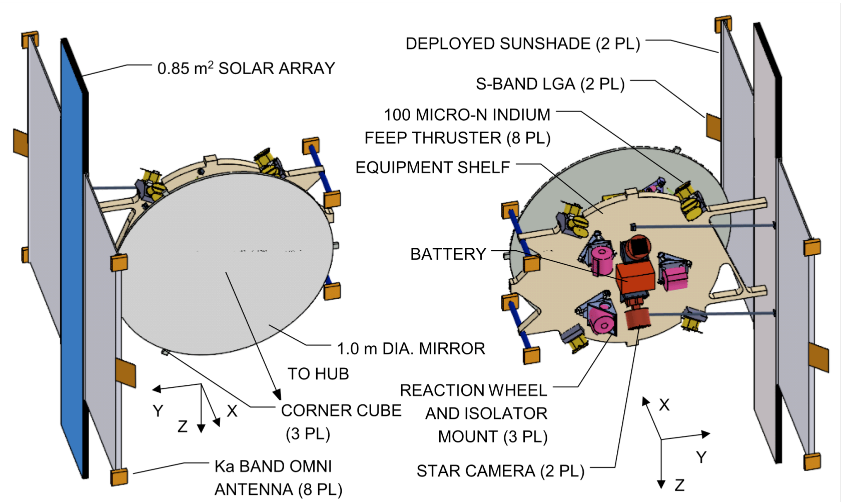
Principal Elements of SI Hub



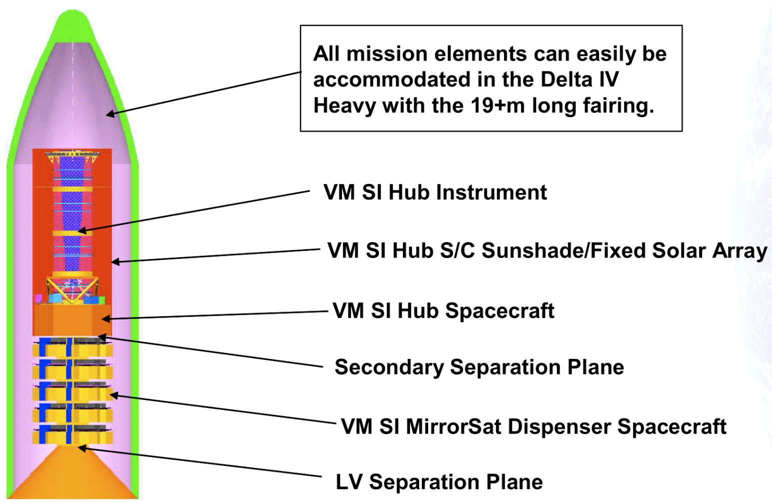
Mirrorsats: Original IMDC Concept



Mirrorsats: BATC (Lightweight) Option

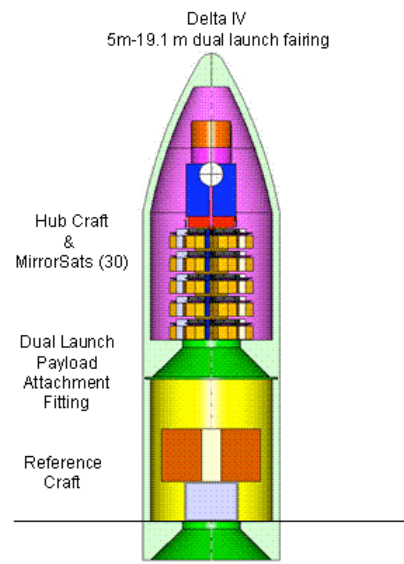


There are several viable launch options for designs with 1-meter array elements (the baseline VM design)

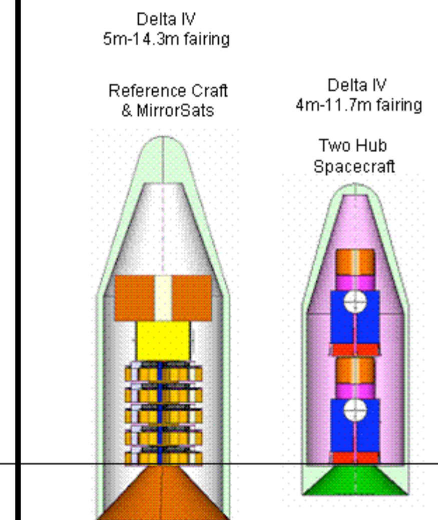


Delta IV Heavy / 19+m fairing:
- 1 hub
- no reference craft
- 30 1-m primary array elements

Delta IV/5x19.1m:
- 1 hub
- 1 reference craft
- 30 1-m primaries



Two Delta IVs:
- 2 hubs
- 1 reference craft
- 30 1-m primaries

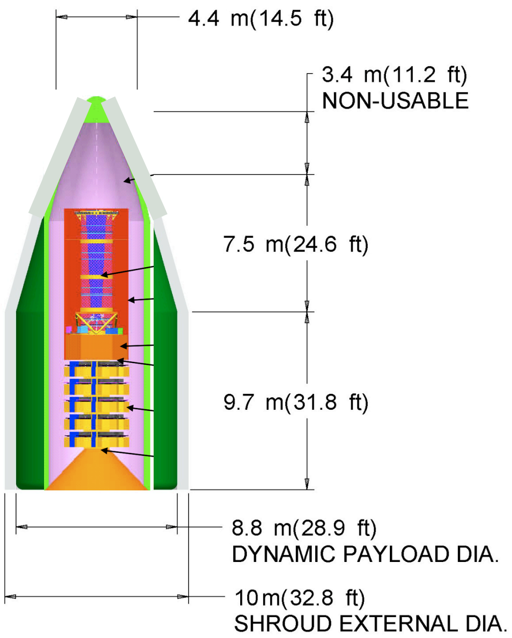


But these options do not support launch of a system with larger than 1-m diameter primary array elements: *this is where the Ares V can help*

(Also: Designs with a reference craft and 2 hubs require 2 launches without Ares V)

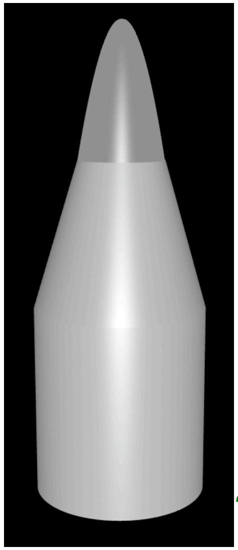
Value to SI of the Ares V (or other similar large fairing vehicle ~Atlas V HLV)

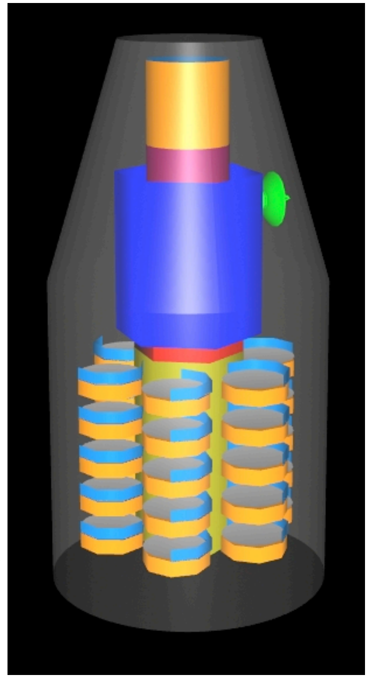
- Ares V (its larger fairing volume) enables inclusion of larger primary array elements
  - VM design has 30 mirrors, each 1m in diameter
  - Larger diameters are desirable for improved sensitivity, but do not fit in 1-2 Delta IV launches
  - With Ares V: 30 x 2m (and larger!) are feasible
    - dramatically increases the sensitivity and science productivity of the observatory, especially for the fainter, extra-galactic sources (e.g., AGN, Quasars, Black Hole environments, etc.)
      - provides equivalent of an 11m diameter monolith in "light-bucket" mode (4x more light than 1m mirrors, nearly 20x light gathering capability of HST)
      - enables much faster asteroseismic observations - shortens the period needed to obtain the million counts needed for the modal studies from 1 month to about 1 week, enabling more stars to be studied in this manner to reveal internal structure and flows
- Ares V may enable launch on a single vehicle of designs which include:
  - more than 1 hub (strongly desired for operational efficiency and redundancy)
  - a reference metrology/pointing control spacecraft

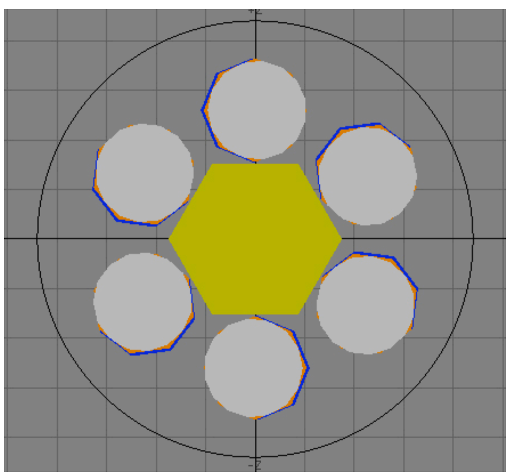


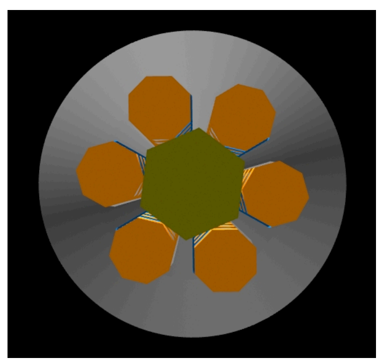
Delta 1m launch configuration fits well inside Ares V shroud!

Packaging SI in an Ares V shroud





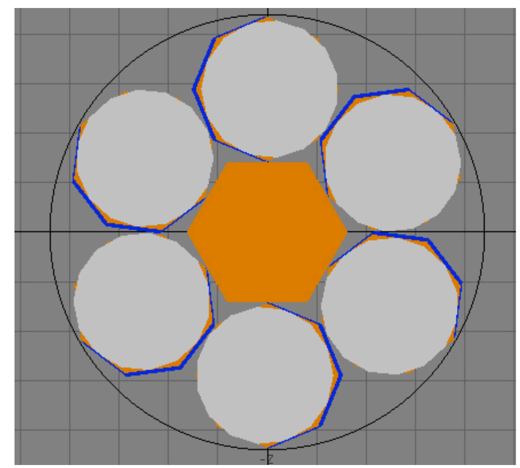




Mirrors 2m in diameter
Dispenser Hexagon 1.75m on a side

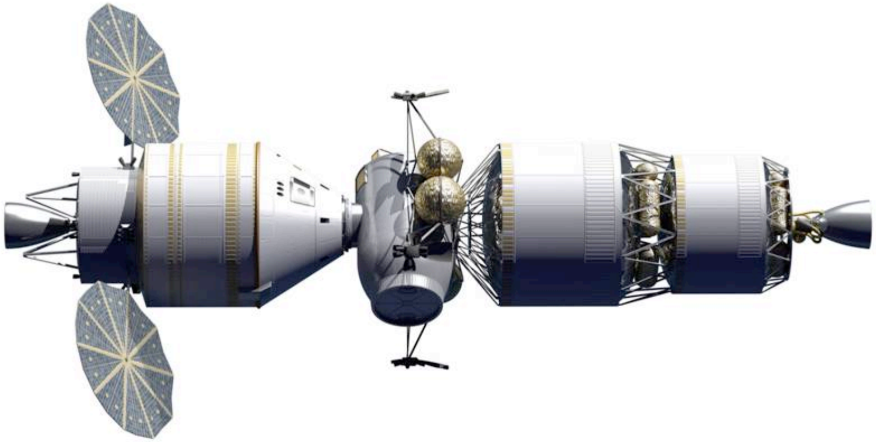
Ares V shroud assumed to have 8.8m inner diameter, 10.0m outer diameter

Largest Mirror Size accommodated by this shroud is 2.75m in diameter (with Hexagon 1.63m on a side)

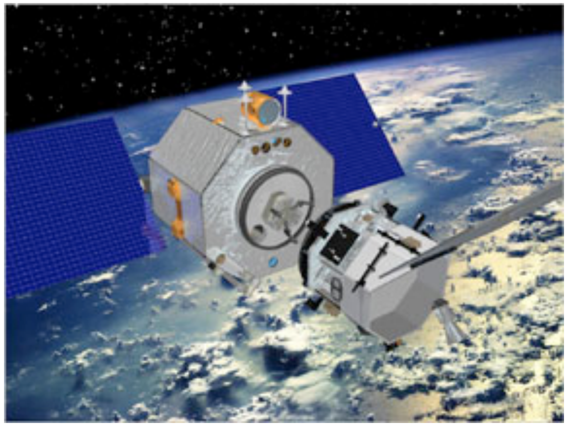


Value of In-Situ Servicing to SI

- SI can benefit significantly if elements can be serviced during extended operations (re-fueled, fixed, replaced), perhaps by humans in the Orion vehicle, or by robotic means…



LSAM L1 Stack
(Orion/CEV mated to a crew module)
http://www.futureinspaceoperations.com/



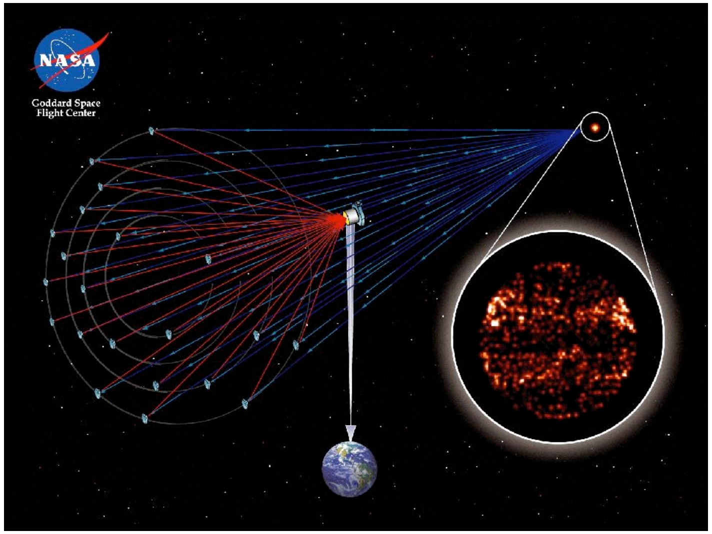
Orbital Express has demonstrated feasibility of autonomous (robotic) on-orbit refueling and reconfiguration:
http://www.darpa.mil/orbitalexpress/

Summary of Major Benefits to SI from use of Constellation Architecture

- Ares V and its larger fairing volume (*or similar proposed HLV versions of the Delta or Atlas vehicles*) enables inclusion of larger primary mirror array elements
  - VM design has 30 mirrors 1m in diameter
  - Larger diameters are desirable for improved sensitivity, but do not fit in one or two launches of a Delta IV
  - With Ares V: 30 x 2m (and larger!) are feasible
  - **Increases sensitivity of observatory and dramatically increases the science productivity, especially for fainter, extra-galactic sources (AGN, Quasars, Black Hole environments, etc.) and shortens the asteroseismic campaigns measuring stellar internal structure and flows from ~1 month to ~1 week.**
- Ares V may enable launch on a single vehicle of designs which include more than 1 hub (strongly desired for operational efficiency and redundancy) and/or a reference metrology/pointing control spacecraft
- SI can also benefit significantly if elements can be serviced during its long operational life (elements re-fueled, fixed, replaced) – a possible role for the manned Orion/CEV or robots deployed by an Ares launch

Stellar Imager (SI): Summary

- UV-Optical Interferometer to provide 0.1 mas spectral imaging of
  - magnetic field structures that govern: formation of stars & planetary systems, habitability of planets, space weather, transport processes on many scales in Universe
- A “Flagship” (Vision) mission in the NASA 2005 Heliophysics Roadmap
- A candidate for the UVOI in the 2006 Astronomy & Physics Div. Science Plan
- Mission Concept
  - 30 “mirrorsats” formation-flying with beam combining hub
  - Launch ~ 2024, to Sun-Earth L_2
  - baselines ~ 100 - 1000 m
  - Mission duration: ~10 years



Prime Science Goals

Understand the Role of Magnetism in the Universe and thereby *revolutionize our understanding* of:

Solar/Stellar Magnetic Activity and their impact on Space Weather, Planetary Climates, and Life

Magnetic and Accretion Processes and their roles in the Origin & Evolution of Structure and in the Transport of Matter throughout the Universe

The close-in structure of Active Galactic Nuclei (AGN) and Quasars, and their winds

Exo-Solar Planet Transits and Disks

http://hires.gsfc.nasa.gov/si/

Additional Information and Alternative Illustrations

SI Requirements Flow Down

Science Goals

Solar/Stellar Magnetic Activity
-Understand the dynamo process responsible for magnetic activity
-Enable improved forecasting of solar/stellar magnetic activity on time scales of days to centuries
-Understand the impact of stellar magnetic activity on planetary climates and on the origin and continued existence of life

Magnetic Accretion Processes
-Understand accretion mechanisms in sources ranging from planet-forming systems to black holes
-Understand the dynamical flow of material and the role of accretion in evolution, structure, and transport of matter in complex interacting systems

AGN Structure
-Understand the close-in structure of AGN including jet forming regions, winds and transition regions between Broad & Narrow Line Emitting Regions.

Dynamic imaging of Universe at ultra-high resolution - understand the dynamical structure and physical processes in many currently unresolved sources, e.g. AGN, SN, PN, Interacting binaries, stellar winds and pulsations, forming-stars and disks regions, evolved stars.

→

Data Required

Examples for solar/stellar targets:
-Empirical constraints to refine dynamo models (e.g. for a solar-type star at 4pc)
-Observations of spatial and temporal stellar surface patterns covering a broad range of magnetic activity levels
-Measurement of internal stellar structure and rotation

⇨ UV (1550 Å, 2800 Å) images with 1000 total resolution elements taken with modest integration times (~hours for dwarfs to days for giants)

⇨ Optical Asteroseismology with 30-100 total resolution elements over a stellar disk to measure non-radial resonant waves [integration times - minutes (dwarfs) to hours (giants)]

Examples for non-stellar targets:
-Measurement of sizes/geometries of BLRs, NLRs and opening angles in AGN; Spectral images of accretion processes in planet-forming regions, interacting binaries, BH environments;
-Dynamic imaging of jet-forming regions and evolving jets, e.g.in AGN, YSOs, PN, SN, interacting binaries

⇨ ~0.1 milliarcsecond imaging with spectral information (R>100) over the 1200 – 6600 A range to provide time-lapse images with dozens of resolution elements

****Mission lifetime of 5 yr (10 yr goal) needed to cover significant fraction of stellar activity cycles***

→

Measurements Req.

Angular Resolution :
0.1 mas @ 2000 Å
Spectral Range
1200 – 6600 Å
Field of View
~ 4 mas minimum
Flux Threshold at 1550 Å
5×10^{-14} ergs/cm^2/s
Observations
-several dozen solar-type stars observed repeatedly over mission lifetime (MLT)
-month-long seismology campaigns on select targets
-a sample of extragalactic & galactic sources (e.g. AGN SN, PN, stars, planet forming regions, binaries) observed several times during the MLT

↓

Engineering Implications

Baselines from 100 to 1000m

~30 primary UV-quality mirrors of > 1 meter diameter

Fizeau Beam combination

Path Length Control to 3 nm

Aspect Control to 30 μas

Orientation +/-20 deg to orthogonal to Sun

→

Key Technologies

-precision metrology and formation-flying

-***wavefront sensing and closed-loop control*** of many-element optical systems

-***deployment/initial positioning*** of elements in large arrays

-***metrology/autonomous nm-level control*** of many-element formations over kms

-variable, non-condensing, continuous μ-Newton thrusters

-***light-weight UV quality spherical mirrors*** with km-long radii of curvature

-***larger format energy resolving detectors*** with finer energy resolution (R=100) or a Spatial Frequency Remapper beam combiner to enable spectral dispersion of each beam

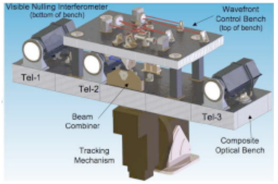
-methodologies for ground-based integration and test of distributed s/c systems

-mass-production of "mirrorsat" spacecraft

Feasibility of Interferometry from Space

- SI is part of a natural evolution from current ground-based interferometers and testbeds to a space-based system (see next page)
- Feasibility of interferometry demonstrated by large variety of successful ground-based interferometers (e.g., CHARA, COAST, NPOI, and VLTI)
  - Their performance, and that of space-based interferometers, can be improved simply by increasing # of elements, as has been done for radio facilities
- Space provide better environment
  - Not looking through an atmosphere, which on the ground limits spatial and temporal coherence (aperture size and integration time) of incoming wavefront
  - No need for large and complicated delay lines for off-axis obs.
  - Wavelengths not available from ground can be accessed
- A simple imaging interferometer, like SI, is a logical first "large baseline, space-based" interferometer
  - it is easier than an astrometric mission like SIM, since its light-path delay tolerance is ~2 orders of mag less than SIM's $\lambda/1000$ level
  - It is easier than TPF-I-like missions aimed at planet detection via nulling the central star and requiring a fringe contrast ~0.99999 and having error requirements ~10000x more severe than SI with its 0.9 fringe contrast requirement
- A small-baseline space interferometer with just a few primary mirrors (e.g., FKSI or Pegase) would be an ideal bridge from the ground-based to large baseline space-based interferometers

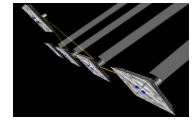
Notional Path for Development of Space Interferometry



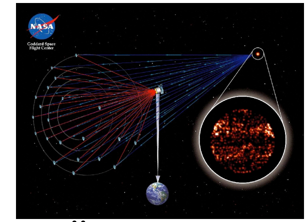
Balloon-Based Missions:
BENI or BETTII

Space Tech. Demos:
ST-9 or Proba-3

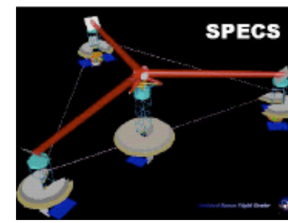
Planet Finders:
SIM & TPF



Large Strategic ("Vision") Imaging Interferometry Space Missions



Stellar Imager
UV-Opt./Magnetic Activity

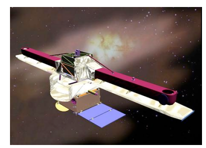


SPIRIT/SPECS
IR "Deep Fields"

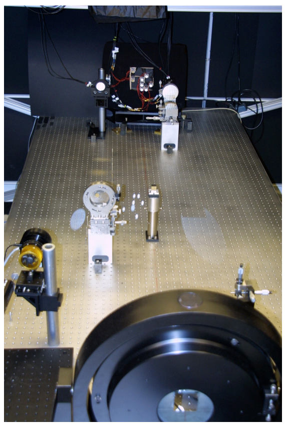
Black Hole Imager
X-ray/BH Event Horizons

Life Finder
Searching for Signs of Life

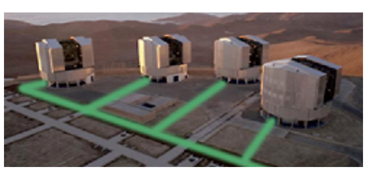
Planet Imager
Terrestrial-Planet Imaging



Smaller Space Interferometers
(e.g., FKSI and/or Pegase)



Ground-Based Testbeds
Wavefront Sensing/Control:
FIT, STAR9, FKSIT
Formation Flying:
SIFFT, FFTB, FCT
Metrology: SAO-TFG



Ground-based interferometers
(Keck,VLTI,LBT, ISI, CHARA, COAST, GI2T, NPOI, MRO)
Giant star imaging, Binary stars

2005 — 2010 — 2015 — 2020 — 2025 +

Enabling Stellar Imager: Technology Investments are Essential

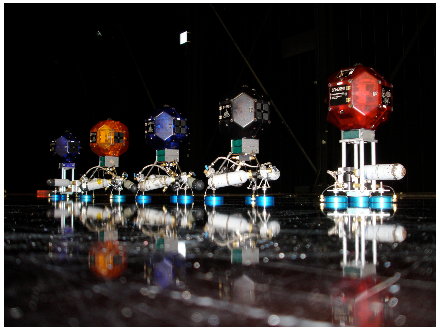
- **formation-flying of ~ 30 spacecraft**
  - deployment and initial positioning of elements in large formations
  - real-time correction and control of formation elements
  - staged-control system (km → cm → nm)
  - aspect control to 10's of micro-arcsec
  - positioning mirror surfaces to 2 nm
  - variable, non-condensing, continuous micro-Newton thrusters
- **precision metrology over multi-km baselines**
  - 2nm if used alone for pathlength control (no wavefront sensing)
  - 0.5 microns if hand-off to wavefront sensing & control for nm-level control
  - multiple modes to cover wide dynamic range
- **wavefront sensing and real-time, autonomous analysis & control**
  - use the science data stream to control nm-level placement of mirrors
- **methodologies for ground-based validation of distributed systems**
- additional challenges (perceived as easier than the above)
  - mass-production of "mirrorsat" spacecraft: cost-effective, high-volume fabrication, integration, & test
  - long mission lifetime requirement
  - light-weight UV quality mirrors with km-long radii of curvature (perhaps deformable UV quality flats)

Addressing the Technical Challenges

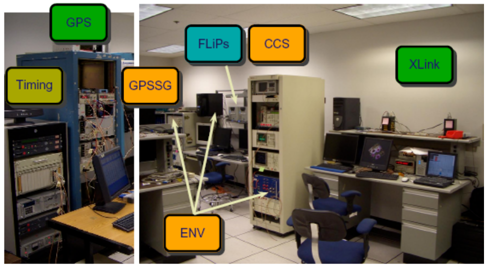
- The technology challenges identified on the previous slide have all been addressed prior to and during the SI Vision Mission (VM) study:
  - in both IMDC and ISAL sessions dedicated to SI development over the period 2001-2005
  - and in other Integrated Design Center studies run as joint efforts with other interferometric design efforts (e.g., a joint study with MAXIM examining and optimizing techniques for aspect control of spacecraft to the 10's of micro-arcsec level).
- Credible and feasible approaches to the successful development of all these technologies were derived during the course of those studies and are documented in the SI VM Final Report.
- A notional "Path for the Development of Space Interferometry" has been developed (see earlier slide)
- In addition, there are a number of ground-based testbeds which are aggressively pursuing the development of these technologies, including the development and assessment of:
  - precision formation flying (PFF) algorithms (SIFFT/SPHERES, FFTB)
  - closed-loop optical control of tip, tilt, and piston of the individual mirrors in a sparse array based on feedback from wavefront analysis of the science data stream (FIT)
  - high-precision metrology (SAO & JPL Testbeds)

GSFC/SI Technology Development Programs

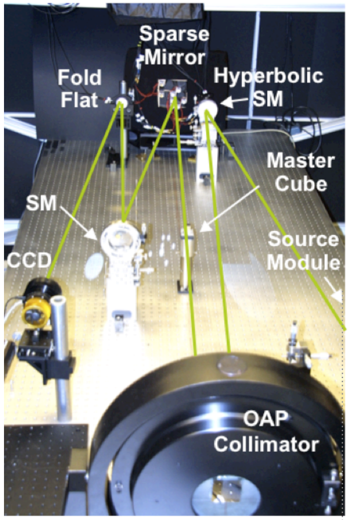
- **GSFC/MSFC/MIT Synthetic Imaging Formation Flying Testbed (SIFFT; Carpenter, Lyon, Stahl, Miller, et al.)**
  - Develop cm-level formation flying algorithms on lab hardware, including Formation Deployment/Maintenance, Reconfiguration, Imaging Maneuvers
  - Uses MIT SPHERES on the MSFC Flat Floor
  - Have demonstrated formation control of 3 floating SPHERES and reconfiguration by rotating/expanding formation



- **GSFC Formation Flying Testbed (FFTB; J. Leitner, E. Stoneking, J. Mitchell, R. Luquette)**
  - Software simulation facility
  - Used to develop & demo deployment of array s/c and multi-stage acquisition of target light from individual mirrors by beam combiner
  - Stoneking simulated all stages of formation acquisition for full-up SI



- **Fizeau Interferometer Testbed (FIT; K. Carpenter, R. Lyon, A. Liu, D. Mozurkewich, P. Petrone, P. Dogoda)**
  - Develop & demo closed-loop, nm-level optical control of a many-element sparse array, ***using wavefront sensing of the science data stream***
  - Develop/assess image synthesis algorithms
  - Develop nulling techniques for Fizeau Interferometers for planet detection/imaging



The Ultimate Goal: develop Staged-Control Methodologies covering over 12 orders of magnitude, from nm to km scales

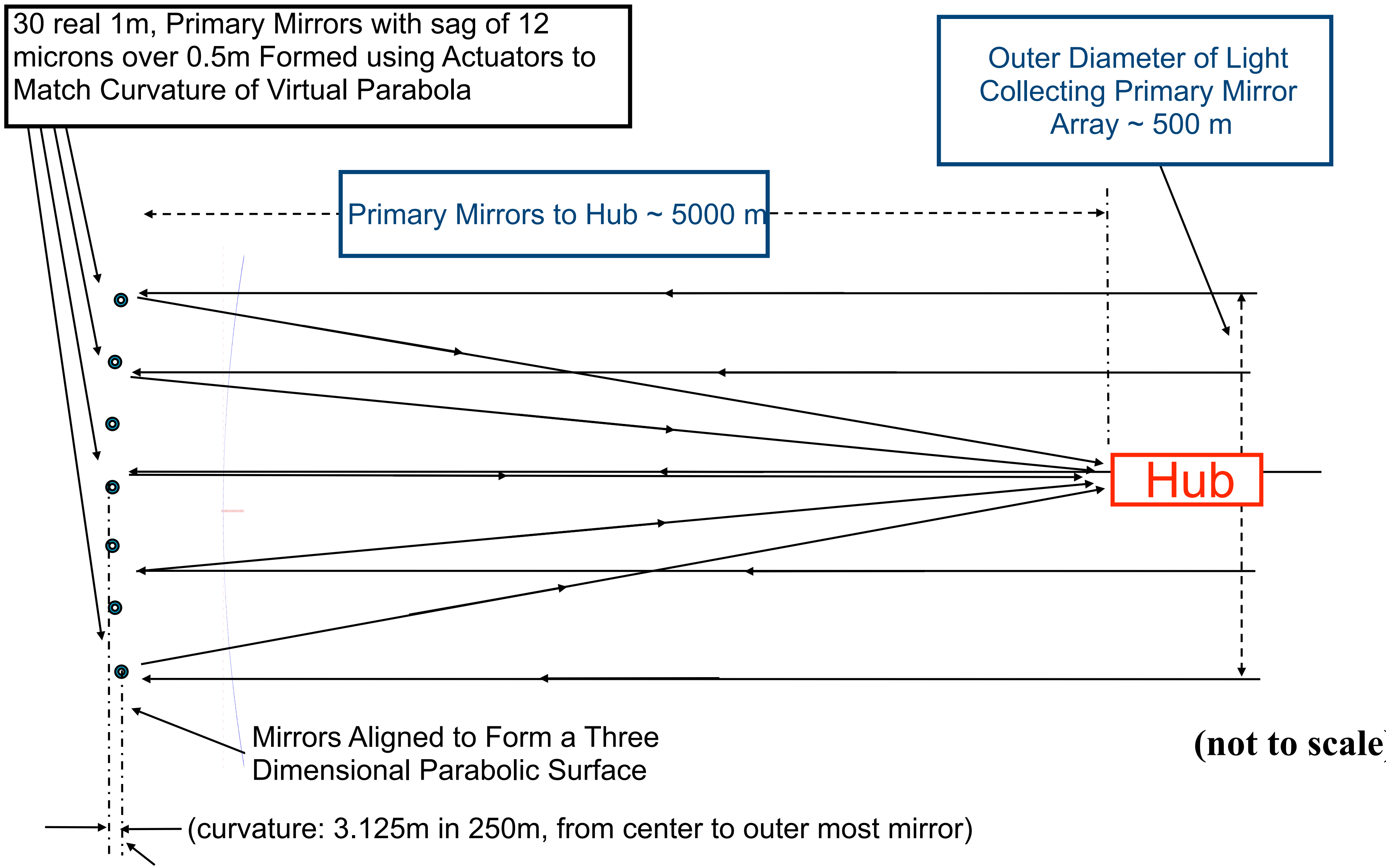
SI Technology Status & Approach to Achieving TRL 6

Required Capability	Now	Figure of Merit Risk Reduction Demo	Full SI	Current TRL	TRL 5/6 Test Requirement
Measure relative position of hub and collecting mirrors	2 cm post-processed	< 2 cm on-board, real-time	cm -> nm on-board	2 cm: 6 < cm: 4	RF or optical channel simulator with high fidelity dynamic simulator and real-time estimation/wavefront sensing
Measure relative bearing angles of collecting mirrors wrt direction to hub	N/A	1 arcmin	0.4 milli-arcsec	4	Prototype integrated into hi-fidelity simulation, w/ real-time estimation and wavefront sensing
Control relative position of collecting mirrors	N/A	10 cm	cm -> nm	4	RF or optical channel simulator with high fidelity dynamic simulator and real-time estimation and control loops wrapped around.
Control relative bearing angles of collecting mirrors wrt direction to hub	N/A	5 arcmin	1.5 mas	2	Prototype integrated into hi-fidelity simulation, with real-time estimation and control loops wrapped around
Formation line-of-sight Control	N/A	10 arcmin	20 μas	3	Interferometric verification
Formation Commanding	Ground	On-Board	On-Board	4	Distributed simulation
Auton. collision avoidance	N	Y	Y	4	High-fidelity simulation
Direct starlight to combiner with little scatter and mass	25 kg/m^2	10 kg/m^2 over 50 cm dia	10kg/m^2 over 1-2m dia.	3	Interferometric and scatterometer measurements of demo mirror/actuators/structure
Control collector mirror surface	5 nm piston; <1 mas tilt			6 for Be mirrors; 2 for polymer	Interferometric verification
Observatory Ground Validation	N/A			2	High-fidelity simulation with measured component performance

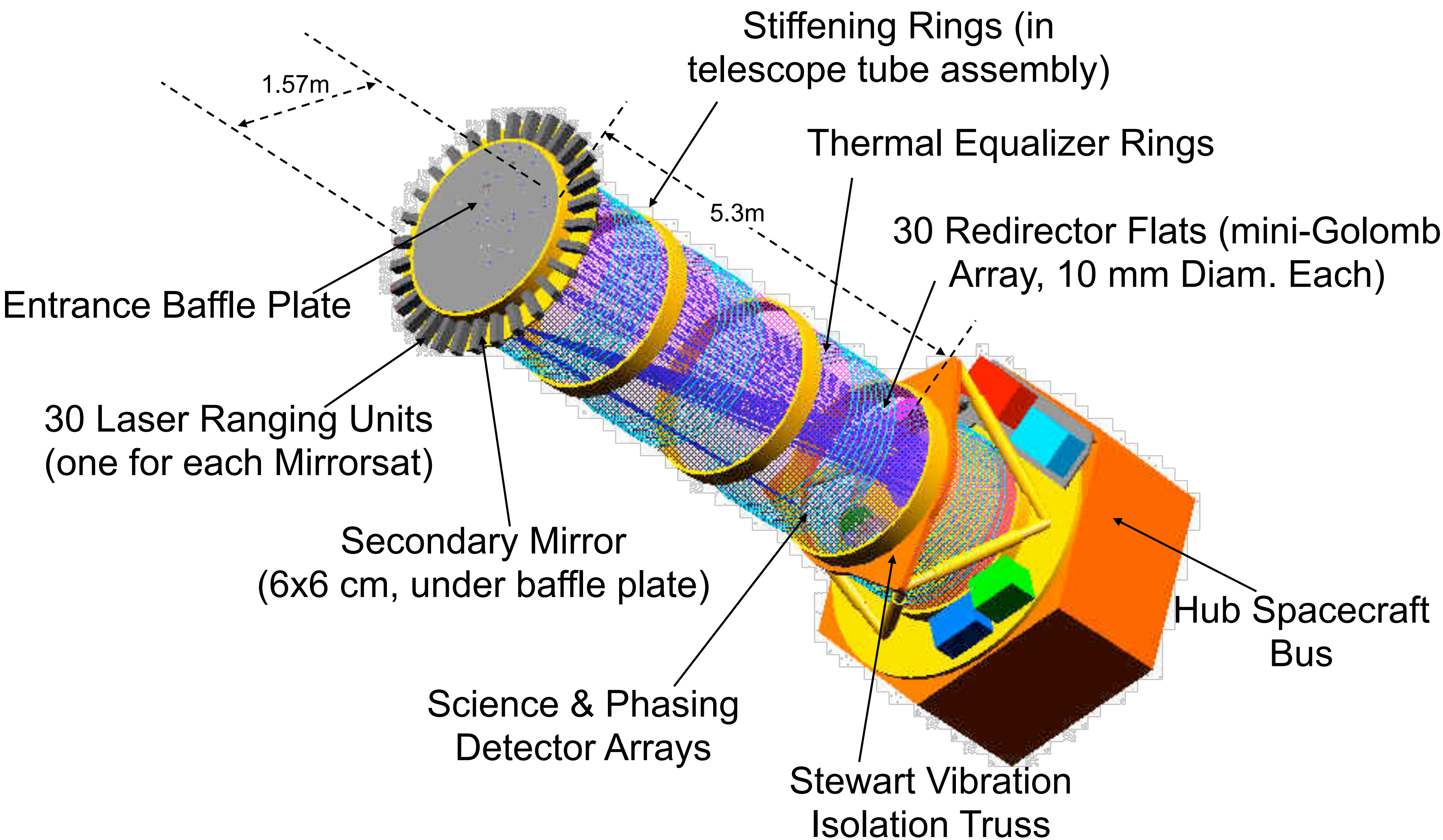
Mission and Performance Parameters

Parameter	Value	Notes
Maximum Baseline (B)	100 – 1000 m (500 m typical)	Outer array diameter
Effective Focal Length	1 – 10 km (5 km typical)	Scales linearly with B
Diameter of Mirrors	1 - 2 m (1 m currently)	Up to 30 mirrors total
λ-Coverage	UV: 1200 – 3200 Å Optical: 3200 – 5000 Å	Wavefront Sensing in optical only
Spectral Resolution	UV: 10 Å (emission lines) UV/Opt: 100 Å (continuum)	
Operational Orbit	Sun-Earth L2 Lissajous, 180 d	200,000x800,000 km
Operational Lifetime	5 yrs (req.) – 10 yrs (goal)	
Accessible Sky	Sun angle: $70^\circ < b < 110^\circ$	Entire sky in 180 d
Hub Dry Mass	1455 kg	Possibly 2 copies
Mirrorsat Dry Mass	65 kg (BATC) - 120 kg (IMDC)	For each of up to 30
Ref. Platform Mass	200 kg	
Total Propellant Mass	750 kg	For operational phase
Angular Resolution	50 mas – 208 mas (@1200–5000Å)	Scales linearly ~ λ/B
Typical total time to image stellar surface	< 5 hours for solar type < 1 day for supergiant	
Imaging time resolution	10 – 30 min (10 min typical)	Surface imaging
Seismology time res.	1 min cadence	Internal structure
# res. pixels on star	~1000 total over disk	Solar type at 4 pc
Minimum FOV	> 4 mas	
Minimum flux detectable at 1550 Å	5.0 x 10^{-14} ergs/cm^2/s integrated over C IV lines	10 Å bandpass
Precision Formation Fly.	s/c control to mm-cm level	
Optical Surfaces Control	Actuated mirrors to mm-nm level	
Phase Corrections	to λ/10 Optical Path Difference	
Aspect Control/Correct.	3 mas for up to 1000 sec	Line of sight maintenance

SI Cross-Sectional Schematic



Principal Elements of SI Hub



Mission Concept Development Team

- Mission concept under development by NASA/GSFC in collaboration with experts from industry, universities, & astronomical institutes:

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Arizona State University	Catholic University of America
Ball Aerospace & Technologies Corp.	Lockheed Martin Adv. Tech. Center
Marshall Space Flight Center	Massachusetts Inst. of Technology
Northrop-Grumman Space Tech.	Seabrook Engineering
Sigma Space Corporation	Smithsonian Astrophysical Observatory
Space Telescope Science Institute	State Univ. of New York/Stonybrook
Stanford University	University of Colorado at Boulder
University of Maryland	University of Texas/Arlington&SanAn.
European Space Agency	College de France
Astrophysical Institute Potsdam	University of Aarhus

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Lockheed Martin Adv. Tech. Center
Massachusetts Inst. of Technology
Seabrook Engineering
Smithsonian Astrophysical Observatory
State Univ. of New York/Stonybrook
University of Colorado at Boulder
University of Texas/Arlington&SanAn.

European Space Agency
Astrophysical Institute Potsdam

College de France
University of Aarhus

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  - K. Carpenter, C. Schrijver, M. Karovska, A. Brown, A. Conti, K. Hartman, S. Kilston, J. Leitner, D. Lakins, A. Lo, R. Lyon, J. Marzouk, D. Miller, D. Mozurkewich, J. Phillips, P. Stahl, F. Walter

- Additional science and technical collaborators from these institutions include:
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- International Partners include:
  - J. Christensen-Dalsgaard, F. Favata, K. Strassmeier, A. Labeyrie